Figure 14: NACA0012 at $M = 1.4$, $\alpha = 0°$ and $Re = 1\,000$: Adapted meshes (left) composed of about 72 000 vertices and associated Mach number field (right) obtained using the goal-oriented viscous error estimate. From top to bottom, global view of the domain, close-up view of the airfoil and zoom in the boundary layer region where the shock-boundary layer interaction occurs.